


EASA	AIRWORTHINESS DIRECTIVE	
	<p>AD No.: 2011-0176</p> <p>Date: 13 September 2011</p> <p>Note: This Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EC) No 216/2008 on behalf of the European Community, its Member States and of the European third countries that participate in the activities of EASA under Article 66 of that Regulation.</p>	
<p>This AD is issued in accordance with EC 1702/2003, Part 21A.3B. In accordance with EC 2042/2003 Annex I, Part M.A.301, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [EC 2042/2003 Annex I, Part M.A.303] or agreed with the Authority of the State of Registry [EC 216/2008, Article 14(4) exemption].</p>		
<p>Type Approval Holder's Name : AIRBUS</p>	<p>Type/Model designation(s) : A320 aeroplanes</p>	
<p>TCDS Number : EASA.A.064</p>		
<p>Foreign AD : Not applicable</p>		
<p>Supersedure : None</p>		
<p>ATA 53</p>	<p>Fuselage – Forward Fuselage Frame (FR) 35 Circumferential Junction – Inspection / Repair</p>	
<p>Manufacturer(s):</p>	<p>Airbus (formerly Airbus Industrie)</p>	
<p>Applicability:</p>	<p>Airbus A320-214 and A320-232 aeroplanes, manufacturer serial numbers 3456, 3503, 3516, 3529, 3591, 3597, 3611, 3631, 3696, 3698, 3714, 3719, 3775, 3777, 3780, 3782, 3786, 3797, 3805, 3812, 3870, 3907, 3909, 3913, 3922, 3929, 3946, 3953, 3975, 3979, 3991, 4010, 4012, 4014, 4027, 4034, 4043, 4046, 4064, 4065, 4084, 4093, 4094 and 4097.</p>	
<p>Reason:</p>	<p>A problem was reported during the installation of upper panels on Frame 35 in Airbus A320 final assembly line. Investigations revealed that medium head fasteners, Part Number (P/N) EN6114V3, were installed in lieu of shear head fasteners, P/N ASNA2657V3 and ASNA2043V3, which were previously used. Installation of these medium head fasteners leads to a deeper countersink in the panel. Fatigue and damage tolerance analyses were performed, the results of which demonstrated that this installation could have a fatigue impact on two rows of fasteners between stringers (STGR) 5 and 6, and indicated the need for a specific inspection in this area.</p> <p>This condition, if not detected and corrected, could impair the structural integrity of the affected aeroplanes.</p> <p>For the reasons described above, this AD requires repetitive special detailed inspections of the affected fasteners and, depending on findings, the accomplishment of associated corrective actions.</p>	
<p>Effective Date:</p>	<p>27 September 2011</p>	

<p>Required Action(s) and Compliance Time(s):</p>	<p>Required as indicated, unless accomplished previously:</p> <ol style="list-style-type: none"> (1) Before the accumulation of 35 900 flight cycles (FC) or 88 100 flight hours (FH), whichever occurs first since aeroplane first flight, and thereafter at intervals not to exceed 28 100 FC or 56 300 FH, whichever occurs first, accomplish a High Frequency Eddy Current (HFEC) inspection of the two rows of six fasteners at FR35 between STGR 5 and STGR 6 at left-hand (LH) and right-hand (RH) side, in accordance with the accomplishment instructions of Airbus Service Bulletin (SB) A320-53-1244. (2) If, during any inspection as required by paragraph (1) of this AD, cracks are detected, before next flight, contact Airbus for approved repair instructions and accomplish those instructions accordingly. (3) Accomplishment of corrective actions as required by paragraph (2) of this AD constitutes terminating action for the repetitive inspections requirements of paragraph (1) of this AD, unless the approved repair instructions provided by Airbus, as required by paragraph (2) of this AD, specify otherwise.
<p>Ref. Publications:</p>	<p>Airbus Service Bulletin A320-53-1244 original issue, dated 17 March 2011.</p> <p>The use of later approved revisions of this document is acceptable for compliance with the requirements of this AD.</p>
<p>Remarks :</p>	<ol style="list-style-type: none"> 1. If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD. 2. This AD was posted on 04 August 2011 as PAD 11-083 for consultation until 01 September 2011. No comments were received during the consultation period. 3. Enquiries regarding this AD should be referred to the Safety Information Section, Executive Directorate, EASA. E-mail ADs@easa.europa.eu. 4. For any question concerning the technical content of the requirements in this AD, please contact: AIRBUS – Airworthiness Office – EIAS, Fax +33 5 61 93 44 51; E-mail: account.airworth-eas@airbus.com.