# Airworthiness Directive

**AD No.: 2020-0089**

**Issued: 17 April 2020**

Note: This Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EU) 2018/1139 on behalf of the European Union, its Member States and of the European third countries that participate in the activities of EASA under Article 129 of that Regulation.

This AD is issued in accordance with Regulation (EU) 748/2012, Part 21.A.3B. In accordance with Regulation (EU) 1321/2014 Annex I, Part M.A.301, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [Regulation (EU) 1321/2014 Annex I, Part M.A.303] or agreed with the Authority of the State of Registry [Regulation (EU) 2018/1139, Article 71 exemption].

**Design Approval Holder's Name:**
AIRBUS

**Type/Model designation(s):**
A380 aeroplanes

Effective Date: 01 May 2020

TCDS Number(s): EASA.A.110

Foreign AD: Not applicable

Supersedure: None

## ATA 54 – Nacelles / Pylons – Inboard Pylon Box at Rib 14 Fasteners – Inspection

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**Manufacturer(s):**
Airbus

**Applicability:**
Airbus A380-841, A380-842 and A380-861 aeroplanes, all manufacturer serial numbers, except those that have embodied Airbus modification 77372 in production.

**Definitions:**
For the purpose of this AD, the following definitions apply:

**The applicable SB:** Airbus Service Bulletin (SB) A380-54-8067 (Engine Alliance GP7200 engines) or SB A380-54-8069 (Rolls-Royce Trent 900 engines), as applicable.

**Affected fasteners:** Fasteners, attaching the lateral panel to Rib 14 at the rear wing to inboard (engine #2 and #3) pylon attachments, installed at locations as indicated in the applicable SB.

**Airbus date of manufacture:** The date of transfer of title (ownership) of the aeroplane upon delivery by Airbus to the first operator.

**Reason:**
During the full scale fatigue test, cracks have been found in the cruciform recess of the head of some inboard (engine #2 and #3) pylon box fasteners, defined in this AD as the affected fasteners.

This condition, if not detected and corrected, could lead, under fatigue effects, to damage on the affected fasteners, possibly resulting in in-flight lateral panel detachment and consequent damage to, and reduced control of, the aeroplane.

To address this potential unsafe condition, Airbus issued the applicable SB to provide instructions to inspect the affected fasteners and the affected fastener holes.

For the reasons described above, this AD requires repetitive inspections of the affected fasteners and holes and, depending on findings, accomplishment of applicable corrective action(s).

## Required Action(s) and Compliance Time(s):

Required as indicated, unless accomplished previously:

**Repetitive Inspection(s):**

(1) Before exceeding 8 800 flight cycles (FC) since Airbus date of manufacture and, thereafter, at intervals not to exceed 8 800 FC, inspect the affected fasteners and holes in accordance with the instructions of the applicable SB.

**Corrective Action(s):**

(2) Depending on findings during each inspection as required by paragraph (1) of this AD, before next flight, accomplish the applicable corrective action(s) and, thereafter, depending on findings, accomplish any applicable follow-on action(s) in accordance with the instructions (see Note 1 of this AD) of the applicable SB, or in accordance with approved instructions provided by Airbus, as applicable.

Note 1: Using Section 1.E, Table 1 and the Flowcharts (for initial and repeat inspection, respectively) as provided in the applicable SB is an acceptable method to determine which action is required and when.

**Terminating Action:**

(3) None.

## Ref. Publications:

Airbus SB A380-54-8067 original issue dated 24 September 2019.

Airbus SB A380-54-8069 original issue dated 24 September 2019.

The use of later approved revisions of the above-mentioned documents is acceptable for compliance with the requirements of this AD.

## Remarks:

1. If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD.

2. This AD was posted on 12 March 2020 as PAD 20-050 for consultation until 09 April 2020. No comments were received during the consultation period.

3. Enquiries regarding this AD should be referred to the EASA Programming and Continued Airworthiness Information Section, Certification Directorate. E-mail: ADs@easa.europa.eu.

4. Information about any failures, malfunctions, defects or other occurrences, which may be similar to the unsafe condition addressed by this AD, and which may occur, or have occurred on a product, part or appliance not affected by this AD, can be reported to the EU aviation safety reporting system.

5. For any question concerning the technical content of the requirements in this AD, please contact: Airbus – IIANA (Airworthiness Office), Telephone: +33 562 110 253, Fax: +33 562 110 307, E-mail: account.airworth-A380@airbus.com.