



Airworthiness Directive

AD No.: 2025-0083

Issued: 14 April 2025

Note: This Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EU) 2018/1139 on behalf of the European Union, its Member States and of the European third countries that participate in the activities of EASA under Article 129 of that Regulation.

This AD is issued in accordance with Regulation (EU) 748/2012, Part 21.A.3B. In accordance with Regulation (EU) 1321/2014 Annex I Part M.A.301, or Annex Vb Part ML.A.301, as applicable, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [Regulation (EU) 1321/2014 Annex I Part M.A.303, or Annex Vb Part ML.A.303, as applicable] or agreed with the Authority of the State of Registry [Regulation (EU) 2018/1139, Article 71 exemption].

Design Approval Holder's Name:

AIRBUS S.A.S.

Type/Model designation(s):

A318, A319, A320 and A321 aeroplanes

Effective Date: 28 April 2025

TCDS Number(s): EASA.A.064

Foreign AD: Not applicable

Supersedure: This AD supersedes EASA AD 2024-0231 dated 04 December 2024.

ATA 25 – Equipment/Furnishings – Cargo Bulkhead Frame 24A Vertical Members – Inspection

Manufacturer(s):

Airbus, formerly Airbus Industrie

Applicability:

Airbus A318-111, A318-112, A318-121, A318-122, A319-111, A319-112, A319-113, A319-114, A319-115, A319-131, A319-132, A319-133, A319-151N, A319-153N, A319-171N, A319-173N, A320-211, A320-212, A320-214, A320-215, A320-216, A320-231, A320-232, A320-233, A320-251N, A320-252N, A320-253N, A320-271N, A320-272N, A320-273N, A321-111, A321-112, A321-131, A321-211, A321-212, A321-213, A321-231, A321-232, A321-251N, A321-251NX, A321-252N, A321-252NX, A321-253N, A321-253NX, A321-253NY, A321-271N, A321-271NX, A321-271NY, A321-272N, and A321-272NX aeroplanes, all manufacturer serial numbers.

Definitions:

For the purpose of this AD, the following definitions apply:

Affected parts: Cargo bulkhead vertical member upper parts and upper fittings located at frame (FR) 24A behind 80VU rack.

The SB: Airbus Service Bulletin (SB) A320-25-1CFU or Airbus SB A320-25-1CFV, as applicable.



Aeroplane date of manufacture: The date of transfer of title (ownership) of the aeroplane upon delivery by Airbus to the first operator, which is referenced in Airbus documentation.

Reason:

During heavy maintenance checks, elongation was found on FR24A cargo panels sub-structure vertical members' assembly upper section (Y-765, Y-254, Y254 and Y765). The affected parts are located behind the 80VU rack.

This condition, if not detected and corrected, could lead to the affected parts hitting the 80VU rack and loss of several computers installed therein, with multiple system failures or (partial) disconnection of systems, which, if occurring during a critical phase of flight, could result in reduced control of the aeroplane.

To address this potential unsafe condition, Airbus issued the SB, as defined in this AD, to provide inspection instructions of the affected parts. Consequently, EASA issued AD 2024-0231 to require a repetitive detailed inspection (DET) of the affected parts and, depending on findings, accomplishment of applicable corrective action(s).

Since that AD was issued, a new aeroplane model (A321-271NY) has been certified, on which affected parts are installed. Furthermore, Airbus updated the naming of the fitting to "upper fittings".

For the reason described above, this AD supersedes EASA AD 2024-0231, extends the Applicability to the A321-271NY aeroplanes and aligns the definition with the Airbus naming to "upper fittings".

For aeroplanes previously affected by EASA AD 2024-0231, this AD retains the requirements of that AD, with no additional action.

Required Action(s) and Compliance Time(s):

Required as indicated by this AD, unless the action(s) required by this AD have been already accomplished:

Inspection(s):

- (1) Before exceeding the threshold as defined in Table 1 of this AD, as applicable, and, thereafter at intervals not exceeding 30 000 flight hours (FH), or 15 000 flight cycles (FC), or 72 months, whichever occurs first, accomplish a DET of each affected part in accordance with the instructions of the SB.



Table 1 - Inspection Thresholds

Threshold (whichever occurs later, A or B)	
A	Before exceeding 30 000 FH, or 15 000 FC since first flight, or 72 months since aeroplane date of manufacture, whichever occurs first
B	Before exceeding 36 months after 18 December 2024 [the effective date of EASA AD 2024-0231]

Corrective Action(s):

- (2) If, during any DET as required by paragraph (1) of this AD, discrepancies are detected, as identified in the SB, before next flight, accomplish the applicable corrective action(s) in accordance with the instructions of the SB.

Terminating Action:

- (3) None.

Reporting:

- (4) If, during any DET as required by paragraph (1) of this AD, discrepancies are detected, as identified in the SB, within 90 days after that DET, or after the effective date of this AD, whichever occurs later, report the inspection results to Airbus. Using the inspection report attached to the SB is acceptable to comply with this requirement.

Ref. Publications:

Airbus SB A320-25-1CFU original issue dated 26 September 2024.

Airbus SB A320-25-1CFV original issue dated 26 September 2024.

The use of later approved revisions of the above-mentioned documents is acceptable for compliance with the requirements of this AD.

Remarks:

1. If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD.
2. Based on the required actions and the compliance time, EASA have decided to issue a Final AD with Request for Comments, postponing the public consultation process until after publication. All interested persons may send their comments, referencing the AD Number, to the E-mail address specified in below Remark 3, prior to 12 May 2025. Only if any comment is received during the consultation period, a Comment Response Document will be published in the [EASA Safety Publications Tool](#), in a compressed ('zipped') file, attached to the record for this AD.
3. Enquiries regarding this AD should be referred to the EASA Safety Information Section, Certification Directorate. E-mail: ADs@easa.europa.eu.



4. Information about any failures, malfunctions, defects or other occurrences, which may be similar to the unsafe condition addressed by this AD, and which may occur, or have occurred on a product, part or appliance not affected by this AD, can be reported to the [EU aviation safety reporting system](#). This may include reporting on the same or similar components, other than those covered by the design to which this AD applies, if the same unsafe condition can exist or may develop on an aircraft with those components installed. Such components may be installed under an FAA Parts Manufacturer Approval (PMA), Supplemental Type Certificate (STC) or other modification.
5. For any question concerning the technical content of the requirements in this AD, please contact: AIRBUS – Airworthiness Office – 1IASA; E-mail: account.airworth-eas@airbus.com.

