



Airworthiness Directive

AD No.: 2024-0124

Issued: 01 July 2024

Note: This Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EU) 2018/1139 on behalf of the European Union, its Member States and of the European third countries that participate in the activities of EASA under Article 129 of that Regulation.

This AD is issued in accordance with Regulation (EU) 748/2012, Part 21.A.3B. In accordance with Regulation (EU) 1321/2014 Annex I Part M.A.301, or Annex Vb Part ML.A.301, as applicable, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [Regulation (EU) 1321/2014 Annex I Part M.A.303, or Annex Vb Part ML.A.303, as applicable] or agreed with the Authority of the State of Registry [Regulation (EU) 2018/1139, Article 71 exemption].

Design Approval Holder's Name:

ATR-GIE AVIONS DE TRANSPORT REGIONAL

Type/Model designation(s):

ATR 72 aeroplanes

Effective Date: 15 July 2024

TCDS Number(s): EASA.A.084

Foreign AD: Not applicable

Supersedure: None

ATA 32 – Landing Gear – Main Landing Gear Hinge Pins – Replacement

Manufacturer(s):

ATR-GIE Avions de Transport Régional, formerly EADS ATR - Alenia, Aerospatiale Matra ATR - ALENIA, Aerospatiale - Alenia, Aerospatiale - Aeritalia

Applicability:

ATR 72-101, ATR72-102, ATR 72-201, ATR 72-202, ATR 72-211, ATR 72-212 and ATR 72-212A aeroplanes, all manufacturer serial numbers.

Definitions:

For the purpose of this AD, the following definitions apply:

The SB: SAFRAN Landing Systems (SLS) Service Bulletin (SB) 631-32-295.

Affected part: Any main landing gear (MLG) hinge pin, having Part Number (P/N) D61000, and serial number (s/n) 14VD48457005, 14VD48457054, 14VD48457058, 14VD48457069, 14VD48457073 or 14VD48457090.

Serviceable part: Any MLG hinge pin, eligible for installation in accordance with applicable ATR instructions, that is not an affected part.



Groups: Group 1 aeroplanes are those that have an affected part installed.
Group 2 aeroplanes are those that do not have an affected part installed.

Reason:

Occurrences of ruptured MLG rear hinge pin, P/N D61000, have been reported. Investigation has been performed by SLS on all hinge pins batches. The investigation results revealed that a batch of 6 pins was subjected to a non-detected thermal abuse in production during grinding process.

This condition, if not corrected, could lead to structural failure and consequent collapse of the MLG, possibly resulting in damage to the aeroplane and injury to the occupants.

To address this potential unsafe condition, SLS issued the SB, providing instructions to replace the affected parts.

For the reasons described above, this AD requires replacement of the affected parts with serviceable parts and prohibits (re)installation of affected part(s) on any aeroplane.

Required Action(s) and Compliance Time(s):

Required as indicated by this AD, unless the action(s) required by this AD have been already accomplished:

Replacement:

- (1) For Group 1 aeroplanes: Within 6 months after the effective date of this AD, replace each affected part with a serviceable part in accordance with the instructions of the SB.

Alternative Method:

- (2) Replacement on an aeroplane of an MLG equipped with an affected part with an MLG having a serviceable part installed is an alternative acceptable method to comply with the requirement of paragraph (1) of this AD for that aeroplane.

Part(s) Installation:

- (3) For Group 1 and Group 2 aeroplanes: From the effective date of this AD, do not install an affected part on any aeroplane or MLG assembly having an affected part installed.

Ref. Publications:

SLS SB 631-32-295 original issue dated 05 March 2024.

The use of later approved revisions of the above-mentioned document is acceptable for compliance with the requirements of this AD.

Remarks:

1. If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD.
2. This AD was posted on 23 May 2024 as PAD 24-056 for consultation until 20 June 2024. The Comment Response Document can be found in the [EASA Safety Publications Tool](#), in the compressed ('zipped') file, attached to the record for this AD.



3. Enquiries regarding this AD should be referred to the EASA Safety Information Section, Certification Directorate. E-mail: ADs@easa.europa.eu.
4. Information about any failures, malfunctions, defects or other occurrences, which may be similar to the unsafe condition addressed by this AD, and which may occur, or have occurred on a product, part or appliance not affected by this AD, can be reported to the [EU aviation safety reporting system](#). This may include reporting on the same or similar components, other than those covered by the design to which this AD applies, if the same unsafe condition can exist or may develop on an aircraft with those components installed. Such components may be installed under an FAA Parts Manufacturer Approval (PMA), Supplemental Type Certificate (STC) or other modification.
5. For any question concerning the technical content of the requirements in this AD, please contact: ATR - GIE Avions de Transport Régional, Continued Airworthiness Service, Telephone: +33 (0)5 62 21 62 21, Fax: +33 (0) 5 62 21 67 18; E-mail: continued.airworthiness@atr-aircraft.com.

