



Airworthiness Directive

AD No.: 2024-0043

Issued: 15 February 2024

Note: This Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EU) 2018/1139 on behalf of the European Union, its Member States and of the European third countries that participate in the activities of EASA under Article 129 of that Regulation.

This AD is issued in accordance with Regulation (EU) 748/2012, Part 21.A.3B. In accordance with Regulation (EU) 1321/2014 Annex I Part M.A.301, or Annex Vb Part ML.A.301, as applicable, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [Regulation (EU) 1321/2014 Annex I Part M.A.303, or Annex Vb Part ML.A.303, as applicable] or agreed with the Authority of the State of Registry [Regulation (EU) 2018/1139, Article 71 exemption].

Design Approval Holder's Name:

FOKKER SERVICES B.V.

Type/Model designation(s):

F28 aeroplanes

Effective Date: 22 February 2024

TCDS Number(s): EASA.A.037

Foreign AD: Not applicable

Supersedure: None

ATA 57 – Wings – Outer Wing Stringers and Stringer Joint Pieces – Inspection

Manufacturer(s):

Fokker Aircraft B.V.

Applicability:

F28 Mark 0070 and Mark 0100 aeroplanes, all serial numbers.

Definitions:

For the purpose of this AD, the following definitions apply:

The SB: Fokker Services Service Bulletin (SB) SBF100-57-056.

Affected aeroplanes: Each aeroplane that, on the effective date of this AD, has accumulated 34 000 flight cycles (FC) or more since entry into service (EIS).

Reason:

Occurrences were reported where, during maintenance of F28 Mark 100 aeroplanes, cracks were found in lower skin stringers at the outboard side of RIB 4700 in the left-hand outer wing of two aeroplanes. Those cracks were detected on aeroplanes which had accumulated a number of FC and/or flight hours which is significantly lower than the threshold for initial inspection of that area as required in the current Airworthiness Limitation Specifications for F28 Mark 0070/0100 aeroplanes.



This condition, if not detected and corrected, could lead to reduced structural integrity of the wing structure, possibly impacting its ability to withstand flight loads.

To address this potential unsafe condition, Fokker Services published the SB, as defined in this AD, to provide inspection instructions and to gather evidence of the number and nature of additional cracks, for further assessment.

For the reason described above, this AD requires, for affected aeroplanes as defined in this AD, a one-time Special Detailed Inspection (SDI), using high frequency eddy current (HFEC) technics, of the stringers on the outboard side of RIB 4700 and of the stringer joint pieces on the inboard and outboard side of RIB 4700 and, depending on findings, accomplishment of applicable corrective action(s). This AD also requires reporting of the inspection results.

This AD is considered an interim action and further AD action may follow.

Required Action(s) and Compliance Time(s):

Required as indicated by this AD, unless the action(s) required by this AD have been already accomplished:

Inspection(s):

- (1) Within the compliance time specified in Table 1 of this AD, depending on the number of FC accumulated by the aeroplane, accomplish an SDI, using HFEC technics, on all the stringers on the outboard side of RIB 4700, and on all the stringer joint pieces on the inboard and outboard side of RIB 4700, of both, the left-hand and right-hand outer wings in accordance with the instructions in the SB.

Table 1 – Compliance Time for the HFEC Inspection

Accumulated FC (see Note 1 of this AD)	Compliance Time A or B, whichever occurs first (as applicable) after the effective date of this AD	
	A	B
45 000 FC or more	A	Within 375 FC
	B	Within 3 months
40 000 FC or more, but less than 45 000 FC	A	Within 1 500 FC
	B	Within 12 months
34 000 FC or more, but less than 40 000 FC	A	Within 3 000 FC
	B	Within 24 months

Note 1: The Accumulated FC indicated in the left column of Table 1 of this AD are those accumulated on the effective date of this AD by an affected aeroplane since EIS.



Corrective Action(s):

- (2) If, during the inspection as required by paragraph (1) of this AD, crack(s) are found, before next flight, contact Fokker Services for approved repair instructions and, within the compliance time(s) indicated therein, accomplish those instructions accordingly.

Reporting:

- (3) Within 15 days after the inspection as required by paragraph (1) of this AD, report the results (including no findings) to Fokker Services. This can be accomplished in accordance with the instructions (reporting sheet) of the SB.

Ref. Publications:

Fokker Services SB SBF100-57-056 original issue dated 15 January 2024.

The use of later approved revisions of the above-mentioned document is acceptable for compliance with the requirements of this AD.

Remarks:

1. If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD.
2. Based on the required actions and the compliance time, EASA have decided to issue a Final AD with Request for Comments, postponing the public consultation process until after publication.
3. Enquiries regarding this AD should be referred to the EASA Safety Information Section, Certification Directorate. E-mail: ADs@easa.europa.eu.
4. Information about any failures, malfunctions, defects or other occurrences, which may be similar to the unsafe condition addressed by this AD, and which may occur, or have occurred on a product, part or appliance not affected by this AD, can be reported to the [EU aviation safety reporting system](#). This may include reporting on the same or similar components, other than those covered by the design to which this AD applies, if the same unsafe condition can exist or may develop on an aircraft with those components installed. Such components may be installed under an FAA Parts Manufacturer Approval (PMA), Supplemental Type Certificate (STC) or other modification.
5. For any question concerning the technical content of the requirements in this AD, please contact: Fokker Services B.V., Technical Services Dept., P.O. Box 1357, 2130 EL, Hoofddorp, The Netherlands; Telephone +31-88-6280-350 or Fax +31-88-6280-111, or E-mail: technicalservices@fokkerservices.com.
The referenced publication can be downloaded from www.myfokkerfleet.com.

