



## Emergency Airworthiness Directive

**AD No.:** 2023-0219-E

**Issued:** 19 December 2023

Note: This Emergency Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EU) 2018/1139 on behalf of the European Union, its Member States and of the European third countries that participate in the activities of EASA under Article 129 of that Regulation.

This AD is issued in accordance with Regulation (EU) 748/2012, Part 21.A.3B. In accordance with Regulation (EU) 1321/2014 Annex I, Part M.A.301, or Annex Vb Part ML.A.301, as applicable, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [Regulation (EU) 1321/2014 Annex I Part M.A.303, or Annex Vb Part ML.A.303, as applicable] or agreed with the Authority of the State of Registry [Regulation (EU) 2018/1139, Article 71 exemption].

**Design Approval Holder's Name:**

PILATUS AIRCRAFT Ltd

**Type/Model designation(s):**

PC-24 aeroplanes

**Effective Date:** 21 December 2023

**TCDS Number(s):** EASA.A.594

**Foreign AD:** Not applicable

**Supersedure:** None

### ATA 27 – Flight Controls – Rudder Trim Tab Installation / Threaded Bolts – Replacement

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**Manufacturer(s):**

Pilatus Aircraft Ltd (Pilatus)

**Applicability:**

PC-24 aeroplanes, manufacturers serial numbers (MSN) 101 and up.

**Definitions:**

For the purpose of this AD, the following definitions apply:

**The SB:** Pilatus PC-24 Service Bulletin (SB) 27-009.

**Affected part:** Threaded titanium bolts, having Part Number (P/N) 527.20.24.489, installed on the rudder trim tab short control rods.

**Serviceable part:** Any threaded titanium bolt, having P/N 527.20.24.489, which is new (not previously installed).



**Reason:**

It has been determined that the titanium threaded bolts at the forward end of the short rudder trim tab actuating rods may be subject to unexpectedly high oscillating loads due to aerodynamic forces acting on the rudder trim tab.

This condition, if not corrected, can lead to failure of an affected part with consequent damage to the rudder and rudder trim tab, possibly resulting in loss of rudder control, and reduced or loss of control of the aeroplane.

To address this potential unsafe condition, pending the development and certification of a new design installation, Pilatus issued the SB, providing instructions to replace the affected parts, and for a one-time inspection.

For the reason described above, this AD requires periodical replacement of affected parts, a one-time inspection of the rudder counterbalance arm and other elements of the rudder trim tab installation and, depending on findings, accomplishment of applicable corrective action(s).

This AD is considered to be an interim action and further AD action may follow.

**Required Action(s) and Compliance Time(s):**

Required as indicated, unless accomplished previously:

**Replacement:**

- (1) Within 10 flight hours (FH) after the effective date of this AD, or within 10 FH after an affected part exceeds 300 FH since first installation on an aeroplane, whichever occurs later, replace that affected part with a serviceable part in accordance with the instructions of the SB (see Note 1 of this AD).

Note 1: If the FH accumulated by an affected part are unknown, the FH accumulated by the aeroplane since first flight may be used instead.

- (2) After the replacement as required by paragraph (1) of this AD, at intervals not to exceed 300 FH, replace each affected part with a serviceable part in accordance with the instructions of the SB.

**Inspection:**

- (3) Within 10 FH after accumulating or exceeding 300 FH since aeroplane first flight, or within 10 FH after the effective date of this AD, whichever occurs later, accomplish a one-time inspection of the rudder mass balance arm in accordance with the instructions of the SB.

**Corrective Action(s):**

- (4) If, during the inspection as required by paragraph (3) of this AD, any discrepancy, as defined in the SB, is identified, before next flight, contact Pilatus for applicable corrective action(s) instructions and, within the compliance time specified therein, accomplish those instructions accordingly.



**Part(s) Installation:**

- (5) From the effective date of this AD, it is allowed to install on any aeroplane an affected part, provided it is a serviceable part, as defined in this AD, and that, following installation, it is replaced as required by paragraph (2) of this AD.

**Terminating Action:**

- (6) None.

**Ref. Publications:**

Pilatus PC-24 SB 27-009 original issue dated 18 December 2023.

The use of later approved revisions of the above-mentioned document is acceptable for compliance with the requirements of this AD.

**Remarks:**

1. If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD.
2. The results of the safety assessment have indicated the need for immediate publication and notification, without the full consultation process.
3. Enquiries regarding this AD should be referred to the EASA Safety Information Section, Certification Directorate. E-mail: [ADs@easa.europa.eu](mailto:ADs@easa.europa.eu).
4. Information about any failures, malfunctions, defects or other occurrences, which may be similar to the unsafe condition addressed by this AD, and which may occur, or have occurred on a product, part or appliance not affected by this AD, can be reported to the [EU aviation safety reporting system](#). This may include reporting on the same or similar components, other than those covered by the design to which this AD applies, if the same unsafe condition can exist or may develop on an aircraft with those components installed. Such components may be installed under an FAA Parts Manufacturer Approval (PMA), Supplemental Type Certificate (STC) or other modification.
5. For any question concerning the technical content of the requirements in this AD, please contact: Pilatus Aircraft Ltd, Technical Support, CH-6371 Stans, Switzerland, Telephone: +41 848 24 7 365, E-mail: [techsupport.ch@pilatus-aircraft.com](mailto:techsupport.ch@pilatus-aircraft.com), Website: [www.pilatus-aircraft.com](http://www.pilatus-aircraft.com).

