



## Airworthiness Directive

**AD No.:** 2025-0227

**Issued:** 16 October 2025

Note: This Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EU) 2018/1139 on behalf of the European Union, its Member States and of the European third countries that participate in the activities of EASA under Article 129 of that Regulation.

This AD is issued in accordance with Regulation (EU) 748/2012, Part 21.A.3B. In accordance with Regulation (EU) 1321/2014 Annex I Part M.A.301, or Annex Vb Part ML.A.301, as applicable, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [Regulation (EU) 1321/2014 Annex I Part M.A.303, or Annex Vb Part ML.A.303, as applicable] or agreed with the Authority of the State of Registry [Regulation (EU) 2018/1139, Article 71 exemption].

**Design Approval Holder's Name:**

AIRBUS HELICOPTERS

**Type/Model designation(s):**

EC 175 B helicopters

**Effective Date:** 13 November 2025

**TCDS Number(s):** EASA.R.150

**Foreign AD:** Not applicable

**Supersedure:** This AD supersedes EASA AD 2021-0192 dated 18 August 2021.

### ATA 62 – Main Rotor – Upper and Lower H Legs – Service Life Limit

**Manufacturer(s):**

Airbus Helicopters (AH)

**Applicability:**

EC 175 B helicopters, all manufacturer serial numbers (s/n).

**Definitions:**

For the purpose of this AD, the following definitions apply:

**The ASB:** AH Alert Service Bulletin (ASB) EC175-04A003 Revision 4.

**Affected part:** Main rotor (MR) upper H legs, Part Number (P/N) M623A4201101 and lower H legs, P/N M623A4231101, having an s/n as listed in Appendix Section 4.A or 4.B of the ASB, as applicable.

**Serviceable part:** MR upper H legs and lower H legs that are not an affected part or an affected part that has accumulated less than 10 years since new (affected part's manufacture date, as defined in this AD).

**Groups:** Group 1 helicopters are those that have an affected part installed. Group 2 helicopters are those that do not have an affected part installed.



**Affected part's manufacture date:** The 'manufacture date' identified for each affected part in Appendix Section 4.A or 4.B of the ASB, as applicable.

In case of inconsistencies between the manufacture date as listed in the Appendix of the ASB and the manufacture date as recorded in the Certificate of Conformity (CoC) of the part, reference must be made to the information provided in the ASB.

**Reason:**

It was identified that the upper and lower H legs of the MR are susceptible to crack development, due to ageing phenomenon.

This condition, if not detected and corrected, could lead to the seizure of an MR H leg, possibly resulting in loss of control of the helicopter.

To address this potential unsafe condition, AH issued ASB EC175-04A003 at Revision 3 to provide instructions to record the affected part's manufacture date on the relevant log card and to introduce a retirement life of 10 years. Consequently, EASA published AD 2021-0192 requiring implementation of the applicable Service Life Limit (SLL), in addition to the existing Airworthiness Limitations Section SLL, expressed in flight hours or flight cycles, after which the affected part must be removed from service.

Since that AD was published, it has been determined that some additional affected parts must be added to the list provided in the ASB. In addition, it has been recorded that some affected parts were delivered with a CoC referencing a manufacture date later than the "affected part's date of manufacture", as defined in this AD. Consequently, AH issued the ASB, as defined in this AD, providing an updated list of affected parts.

For the reasons described above, this AD retains the requirements of EASA AD 2021-0192, which is superseded, expands the list of affected parts, and provides additional instructions in the definition of "affected part's manufacture date".

**Required Action(s) and Compliance Time(s):**

Required as indicated by this AD, unless the actions required by this AD have been already accomplished:

**Part Replacements / SLL Implementation:**

- (1) For Group 1 helicopters: Within the compliance time defined in Table 1 of this AD, as applicable, replace each affected part with a serviceable part, as defined in this AD, in accordance with the instructions of Section 3.B of the ASB.

Table 1 – Affected Part Replacement (see Note 1 of this AD)

| Time Accumulated               | Compliance Time                                                                   |
|--------------------------------|-----------------------------------------------------------------------------------|
| Less than 9 years and 9 months | Before exceeding 10 years since the affected part's manufacture date              |
| 9 years and 9 months or more   | Within 3 months after 01 September 2021 [the effective date of EASA AD 2021-0192] |



Note 1: The “time accumulated” specified in Table 1 of this AD is the time accumulated, on 01 September 2021 [the effective date of EASA AD 2021-0192], by the affected part since new (affected part’s manufacture date).

#### **Part(s) Installation:**

- (2) For Group 1 and Group 2 helicopters: From the effective date of this AD, it is allowed to install on any helicopter an affected part, provided it is a serviceable part, as defined in this AD, and that, following installation, that part is replaced as required by this AD.

#### **Ref. Publications:**

AH ASB EC175-04A003 Revision 4 dated 07 October 2025.

The use of later approved revisions of the above-mentioned document is acceptable for compliance with the requirements of this AD.

#### **Remarks:**

1. If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD.
2. Based on the required actions and the compliance time, EASA have decided to issue a Final AD with Request for Comments, postponing the public consultation process until after publication. All interested persons may send their comments, referencing the AD Number, to the E-mail address specified in below Remark 3, prior to 13 November 2025. Only if any comment is received during the consultation period, a Comment Response Document will be published in the [EASA Safety Publications Tool](#), in a compressed (‘zipped’) file, attached to the record for this AD.
3. Enquiries regarding this AD should be referred to the EASA Safety Information Section, Certification Directorate. E-mail: [ADs@easa.europa.eu](mailto:ADs@easa.europa.eu).
4. Information about any failures, malfunctions, defects or other occurrences, which may be similar to the unsafe condition addressed by this AD, and which may occur, or have occurred on a product, part or appliance not affected by this AD, can be reported to the [EU aviation safety reporting system](#). This may include reporting on the same or similar components, other than those covered by the design to which this AD applies, if the same unsafe condition can exist or may develop on an aircraft with those components installed. Such components may be installed under an FAA Parts Manufacturer Approval (PMA), Supplemental Type Certificate (STC) or other modification.
5. For any question concerning the technical content of the requirements in this AD, please contact: Airbus Helicopters (Technical Support), Aéroport de Marseille Provence 13725 Marignane Cedex, France, Telephone +33 (0)4 42 85 97 97, Fax +33 (0)4 42 85 99 66, Web portal: <https://airbusworld.helicopters.airbus.com> / Technical Requests Management, Telephone +33 (0)4 42 85 97 89, or E-mail: [support.technical-airframe.ah@airbus.com](mailto:support.technical-airframe.ah@airbus.com).

