



Airworthiness Directive

AD No.: 2024-0239

Issued: 10 December 2024

Note: This Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EU) 2018/1139 on behalf of the European Union, its Member States and of the European third countries that participate in the activities of EASA under Article 129 of that Regulation.

This AD is issued in accordance with Regulation (EU) 748/2012, Part 21.A.3B. In accordance with Regulation (EU) 1321/2014 Annex I Part M.A.301, or Annex Vb Part ML.A.301, as applicable, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [Regulation (EU) 1321/2014 Annex I Part M.A.303, or Annex Vb Part ML.A.303, as applicable] or agreed with the Authority of the State of Registry [Regulation (EU) 2018/1139, Article 71 exemption].

Design Approval Holder's Name:

Costruzioni Aeronautiche Tecnam S.p.A.

Type/Model designation(s):

P2010 aeroplanes

Effective Date: 24 December 2024

TCDS Number(s): EASA.A.576

Foreign AD: Not applicable

Supersedure: None

ATA 27 – Flight Controls – Rudder Pedals Torque Tube Hinges – Modification

Manufacturer(s):

Costruzioni Aeronautiche Tecnam S.p.A. (Tecnam)

Applicability:

Tecnam P2010 and P2010 TDI aeroplanes, all serial numbers (s/n) up to s/n 317 (inclusive), except those on which Tecnam modification MOD2010/359 has been embodied.

Definitions:

For the purpose of this AD, the following definitions apply:

The SB: Tecnam Service Bulletin (SB) 817-CS Edition (Ed.) 1 Revision (Rev.) 0.

Reason:

An occurrence was reported of disconnection of a rudder pedals torque tube from one of its hinges.

This condition, if not corrected, could cause the rudder pedals assembly to slip out of its housing, possibly resulting in loss of control of the aeroplane.

To address this potential unsafe condition, Tecnam designed the modification MOD2010/359, introducing larger diameter retainer washers on the rudder pedals torque tube hinges, and issued the SB for modification of in-service aeroplanes.



For the reason described above, this AD requires modification of aeroplanes in accordance with the instruction of the SB.

Required Action(s) and Compliance Time(s):

Required as indicated by this AD, unless the actions required by this AD have been already accomplished:

Modification:

- (1) Within 25 flight hours or 30 days, whichever occurs first after the effective date of this AD, modify the aeroplane in accordance with the instructions of the SB.
- (2) Reserved.

Ref. Publications:

Tecnam SB 817-CS original issue (Ed. 1, Rev. 0) dated 06 December 2024.

The use of later approved revisions of the above-mentioned document is acceptable for compliance with the requirements of this AD.

Remarks:

1. If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD.
2. Based on the required actions and the compliance time, EASA have decided to issue a Final AD with Request for Comments, postponing the public consultation process until after publication. All interested persons may send their comments, referencing the AD Number, to the E-mail address specified in below Remark 3, prior to 07 January 2025. Only if any comment is received during the consultation period, a Comment Response Document will be published in the [EASA Safety Publications Tool](#), in a compressed ('zipped') file, attached to the record for this AD.
3. Enquiries regarding this AD should be referred to the EASA Safety Information Section, Certification Directorate. E-mail: ADs@easa.europa.eu.
4. Information about any failures, malfunctions, defects or other occurrences, which may be similar to the unsafe condition addressed by this AD, and which may occur, or have occurred on a product, part or appliance not affected by this AD, can be reported to the [EU aviation safety reporting system](#). This may include reporting on the same or similar components, other than those covered by the design to which this AD applies, if the same unsafe condition can exist or may develop on an aircraft with those components installed. Such components may be installed under an FAA Parts Manufacturer Approval (PMA), Supplemental Type Certificate (STC) or other modification.
5. For any question concerning the technical content of the requirements in this AD, please contact: TECNAM - Airworthiness Office – Email: airworthiness@tecnam.com.

