


<b>EASA</b>	<b>AIRWORTHINESS DIRECTIVE</b>	
	<p><b>AD No.: 2015-0060</b></p> <p><b>Date: 10 April 2015</b></p> <p>Note: This Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EC) No 216/2008 on behalf of the European Community, its Member States and of the European third countries that participate in the activities of EASA under Article 66 of that Regulation.</p>	
<p>This AD is issued in accordance with EU 748/2012, Part 21.A.3B. In accordance with EU 1321/2014 Annex I, Part M.A.301, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [EU 1321/2014 Annex I, Part M.A.303] or agreed with the Authority of the State of Registry [EC 216/2008, Article 14(4) exemption].</p>		
<p><b>Design Approval Holder's Name:</b> PILATUS AIRCRAFT Ltd.</p>	<p><b>Type/Model designation(s):</b> PC-12 aeroplanes</p>	
TCDS Number:	EASA.A.089	
Foreign AD:	Not applicable	
Supersedure:	None	
<b>ATA 27</b>	<b>Flight Controls – Aileron Control System / Tab Counter Balance Weight – Replacement</b>	
Manufacturer(s):	Pilatus Aircraft Ltd.	
Applicability:	<p>PC-12/47 and PC-12/47E aeroplanes, all manufacturer serial numbers (MSN).</p> <p>Note: Aeroplanes from MSN 1521 on are delivered with Pilatus Aircraft Ltd. Service Bulletin (SB) 27-021 already embodied in production.</p>	
Reason:	<p>During a continued airworthiness review, a potential unsafe condition was identified that could result from a disconnected aileron trim tab occurring above an altitude of 10.000 feet.</p> <p>This condition, if not corrected, could lead, in case of a disconnection of an aileron trim tab, to undamped aeroplane vibrations, potentially resulting in structural failure.</p> <p>To address this potential unsafe condition, Pilatus Aircraft Ltd. issued SB No. 27-021 to provide instructions for replacement of the aileron tab counter balance weight.</p> <p>For the reason described above, this AD requires replacement of the aileron tab counter balance weight with a new, slightly heavier, aileron tab counter balance weight.</p>	
Effective Date:	24 April 2015	

<p>Required Action(s) and Compliance Time(s):</p>	<p>Required as indicated, unless accomplished previously:</p> <p>(1) For aeroplanes equipped with a part identified by Part Number (P/N) in Table 1 of this AD, within 12 months after the effective date of this AD, replace the aileron tab counter balance weight and re-identify the aileron trim tab assembly in accordance with the instructions of Pilatus Aircraft Ltd. SB No. 27-021.</p> <p style="text-align: center;">Table 1 – Affected Parts</p> <table border="1" data-bbox="563 434 1460 768"> <thead> <tr> <th>Part Name</th> <th>P/N</th> </tr> </thead> <tbody> <tr> <td rowspan="2">Aileron Trim Tab Assembly</td> <td>527.15.12.037</td> </tr> <tr> <td>527.15.12.038</td> </tr> <tr> <td rowspan="4">Aileron Assembly</td> <td>557.05.12.015</td> </tr> <tr> <td>557.05.12.016</td> </tr> <tr> <td>557.05.12.017</td> </tr> <tr> <td>557.05.12.018</td> </tr> </tbody> </table> <p>(2) Do not install on any aeroplane an aileron trim tab assembly having a P/N as identified in Table 1 of this AD, as follows:</p> <p>(2.1) For an aeroplane that, on the effective date of this AD, has an affected P/N installed: After modification of that aeroplane as required by paragraph (1) of this AD.</p> <p>(2.2) For an aeroplane that, on the effective date of this AD, does not have an affected P/N installed: From the effective date of this AD.</p> <p>(3) From the effective date of this AD, it is allowed to install on an aeroplane an aileron assembly, having a P/N as identified in Table 1 of this AD, provided that it is determined that this does not have an aileron trim tab assembly installed, having a P/N as identified in Table 1 of this AD.</p>	Part Name	P/N	Aileron Trim Tab Assembly	527.15.12.037	527.15.12.038	Aileron Assembly	557.05.12.015	557.05.12.016	557.05.12.017	557.05.12.018
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<p>Ref. Publications:</p>	<p>Pilatus Aircraft Ltd SB No. 27-021 initial issue, dated 20 January 2015.</p> <p>The use of later approved revisions of this document is acceptable for compliance with the requirements of this AD.</p>										
<p>Remarks:</p>	<ol style="list-style-type: none"> <li>1. If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD.</li> <li>2. This AD was posted on 09 March 2015 as PAD 15-024 for consultation until 06 April 2015. No comments were received during the consultation period.</li> <li>3. Enquiries regarding this AD should be referred to the Safety Information Section, Certification Directorate, EASA. E-mail: <a href="mailto:ADs@easa.europa.eu">ADs@easa.europa.eu</a>.</li> <li>4. For any question concerning the technical content of the requirements in this AD, please contact:  PILATUS AIRCRAFT LTD, Customer Support Manager,  CH-6371 STANS, Switzerland.  Tel.: +41 (0)41 619 33 33 Fax: +41 (0)41 619 73 11.  E-mail: <a href="mailto:SupportPC12@pilatus-aircraft.com">SupportPC12@pilatus-aircraft.com</a>.</li> </ol>										