



## Airworthiness Directive

**AD No.:** 2021-0010

**Issued:** 11 January 2021

Note: This Airworthiness Directive (AD) is issued by EASA, acting in accordance with Regulation (EU) 2018/1139 on behalf of the European Union, its Member States and of the European third countries that participate in the activities of EASA under Article 129 of that Regulation.

This AD is issued in accordance with Regulation (EU) 748/2012, Part 21.A.3B. In accordance with Regulation (EU) 1321/2014 Annex I, Part M.A.301, the continuing airworthiness of an aircraft shall be ensured by accomplishing any applicable ADs. Consequently, no person may operate an aircraft to which an AD applies, except in accordance with the requirements of that AD, unless otherwise specified by the Agency [Regulation (EU) 1321/2014 Annex I, Part M.A.303] or agreed with the Authority of the State of Registry [Regulation (EU) 2018/1139, Article 71 exemption].

### Design Approval Holder's Name:

PILATUS AIRCRAFT Ltd

### Type/Model designation(s):

PC-12/47E aeroplanes

**Effective Date:** 25 January 2021

**TCDS Number(s):** EASA.A.089

**Foreign AD:** Not applicable

**Supersedure:** None

## ATA 28 – Fuel – Inward Vent Valve – Modification

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### Manufacturer(s):

Pilatus Aircraft Ltd

### Applicability:

PC-12/47E aeroplanes, manufacturer serial number (MSN) 1720, and MSN 2001 and higher.

### Definitions:

For the purpose of this AD, the following definitions apply:

**The SB:** Pilatus Aircraft PC-12 Service Bulletin (SB) 28-015.

**Affected part:** Inward vent valves, having Part Number 963.04.26.520 and a serial number as listed in section 1.C of the SB.

**Serviceable part:** Any inward vent valve eligible for installation that is not an affected part.

**Groups:** Group 1 aeroplanes are those that have an affected part installed.

Group 2 aeroplanes are those that do not have an affected part installed. Aeroplanes having MSN 2033, 2047 and 2049 or higher, are Group 2, provided it is determined that no affected part has been installed on the aeroplane since its first flight.



**Reason:**

An occurrence was reported where, on the production line, a batch of inward vent valves without a chromate conversion coating on the bonding surface were installed on some PC-12/47E aeroplanes. Such inward vent valves are not in compliance with the latest approved design data.

This condition, if not corrected, could lead to corrosion, consequent degradation of the electrical bonding to Rib 16, and in case of lightning strike, to arcing between the ungrounded equipment and the primary structure, possibly resulting in a fire and reduced control of the aeroplane.

To address this potential unsafe condition, Pilatus issued the SB to provide modification instructions.

For the reason described above, this AD requires modification of each affected part, as defined in this AD. This AD also prohibits (re-)installation of affected parts.

**Required Action(s) and Compliance Time(s):**

Required as indicated, unless accomplished previously:

**Modification:**

- (1) For Group 1 aeroplanes: During the next scheduled inspection, or within 1 200 flight hours or 9 months, whichever occurs first after the effective date of this AD, modify each affected part in accordance with the instructions of section 3.B of the SB.

**Parts Installation:**

- (2) Do not install an affected part on any aeroplane, as required by paragraph (2.1) or (2.2) of this AD, as applicable.
  - (2.1) For Group 1 aeroplanes: After the modification of each affected part on the aeroplane as required by paragraph (1) of this AD.
  - (2.2) For Group 2 aeroplanes: From the effective date of this AD.

**Ref. Publications:**

Pilatus Aircraft PC-12 SB 28-015 original issue dated 12 October 2020.

The use of later approved revisions of the above-mentioned document is acceptable for compliance with the requirements of this AD.

**Remarks:**

1. If requested and appropriately substantiated, EASA can approve Alternative Methods of Compliance for this AD.
2. This AD was posted on 10 December 2020 as PAD 20-195 for consultation until 07 January 2021. No comments were received during the consultation period.
3. Enquiries regarding this AD should be referred to the EASA Safety Information Section, Certification Directorate. E-mail: [ADs@easa.europa.eu](mailto:ADs@easa.europa.eu).



4. Information about any failures, malfunctions, defects or other occurrences, which may be similar to the unsafe condition addressed by this AD, and which may occur, or have occurred on a product, part or appliance not affected by this AD, can be reported to the [EU aviation safety reporting system](#). This may include reporting on the same or similar components, other than those covered by the design to which this AD applies, if the same unsafe condition can exist or may develop on an aircraft with those components installed. Such components may be installed under an FAA Parts Manufacturer Approval (PMA), Supplemental Type Certificate (STC) or other modification.
  
5. For any question concerning the technical content of the requirements in this AD, please contact: Pilatus Aircraft Ltd, Customer Support PC-12, CH-6371 Stans, Switzerland, Telephone: +41 41 619 33 33, Fax: +41 41 619 73 11  
E-mail: [SupportPC12@pilatus-aircraft.com](mailto:SupportPC12@pilatus-aircraft.com), Website: [www.pilatus-aircraft.com](http://www.pilatus-aircraft.com).

