



Lufttüchtigkeitsanweisung (LTA) Consigne de Navigabilité (CN) Direttive sulla Navigabilità (DN) Airworthiness Directive (AD)		FOCA AD HB-2016-001
Inkraftsetzung Mise en vigueur Entrata in vigore Effective Date	17 May 2016	Pilatus – PC-7 FOCA TC/TCDS No: F 56-20

Issue Date: 03 May 2016

ATA Chapter: ATA 53 – Fuselage

Subject: **Fuselage – Main Frames – Frame 11 Fittings – Inspection**

Supersedure / Revised AD(s): Not applicable

Type Certificate Holder's Name: Pilatus Aircraft Ltd.

Manufacturer(s): Pilatus Aircraft Ltd.

Applicability: Model PC-7 aircraft, Manufacturers Serial Numbers (MSN) 101 through MSN 618.

Reason: This Airworthiness Directive (AD) is prompted due to a report of Stress Corrosion Cracking (SCC) on the Main Frame on Frame (FR) 11 left fitting Part Number (P/N) 112.35.07.489 and right fitting P/N 112.35.07.490.

Such a condition, if left uncorrected, could lead to potential loss of the horizontal stabilizer.

In order to correct and control the situation, this AD requires a one-time check to identify the material specification and inspect the affected areas of the airframe that are made of aluminium alloy AA2024-T351. Any structural parts of the aircraft structure found to be cracked must be reported to Pilatus prior to further flight.

Required Action(s) and Compliance Time(s): Required as indicated below, unless already accomplished:

- (1) Within the next 120 days after the effective date of this AD, perform a test to identify the material specification of the left and right fittings P/N 112.35.07.489 and P/N 112.35.07.490 as required by paragraph (§) 3.B. of PILATUS PC-7 Service Bulletin No. 53-013.
- (2) If during the inspection required by § (1) of this AD, fittings made of aluminium alloy AA2124-T851 are found, report the inspection results as required by § 3.D. of PILATUS PC-7 Service Bulletin No. 53-013.
- (3) If during the inspection required by § (1) of this AD, fittings made of aluminium alloy AA2024-T351 are found, before next flight after the effective date of this AD, perform an inspection for cracks as required by § 3.C. of PILATUS PC-7 Service Bulletin No. 53-013 for cracks. No cracks are allowed.

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Note 1: The Aircraft Maintenance Manual (AMM) Temporary Revision (TR) No. 05-34 introduces a scheduled maintenance interval of 12 months to check for cracks at the FR 11 fittings made of aluminum alloy AA2024-T351.

- (4) If during the inspection required by § (3) of this AD, cracks are found, prior further flight, replace the fittings in accordance with the instructions given in PILATUS PC-7 Service Bulletin No. 53-014.

Ref. Publication(s): PILATUS PC-7 Service Bulletin No. 53-013, initial issue.
PILATUS PC-7 Service Bulletin No. 53-014, initial issue.

The use of later approved revisions of this document is acceptable for compliance with the requirements of this AD.

For further information contact: The applicable manufacturer's documentation may be obtained directly from:

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