

Fairchild (Swearingen) SA226 and SA227 Series Aeroplanes

AD/SWSA226/61

Rudder Trim Tab Rod Assembly

2/91 TX

Applicability: All model SA 26, SA226, and SA227 series aircraft.

Requirement: Action in accordance with the technical requirements of FAA Emergency Airworthiness Directive 90-24-03, which requires the following:

Visually inspect the rudder trim tab link assemblies (Part Numbers 27-42025-001 through 27-42025-009 as installed) as follows:

- (1) Remove the fairing strip between the vertical fin and rudder.
- (2) Check each connecting rod end for freedom of movement and corrosion around the bearing as follows:
 - (i) Move the rudder trim system from full left to full right deflection and check for any indications of corrosion or binding in the rod and fittings.
 - (ii) If necessary, remove the bolt connecting the actuator and each rod end and check the bearings for freedom of movement.
 - (iii) Check the bolts connecting the rudder actuator to each rod end to insure each bolt is oriented vertically.
- (3) If either rod end is corroded, prior to further flight replace the affected rod end with a serviceable part.
- (4) If the rudder trim mechanism is incorrectly installed, or if either rod end bearing is binding, prior to further flight replace the affected connecting rod and rod end assembly with serviceable parts.
- (5) If corrosion or binding is not found, reinstall the fairing strip and return the aeroplane to service.

Note 1: Fairchild Aircraft Service Notes 26-SN-061, 226-SN-162, and 227-SN-074 pertain to the subject of this AD.

Note 2: Although not required by this AD, the inspections specified in this AD should be included in the regular aircraft maintenance program.

Compliance: Within 15 hours time in service after 7 December 1990 or prior to 17 December 1990; whichever occurs first.

Background: This AD requires a one-time inspection of the rudder trim tab rod assemblies in all Fairchild SA26, SA226, and SA227 series aircraft. The FAA has been notified of three failures of these assemblies (Part Number 27-42025-001 through 27-42025-009) caused by improper installation in one instance, and corrosion in the other two instances. The actions specified in this AD will preclude failure of the rudder trim tab rod assemblies and the resultant aerodynamic vibration, structural deformation, and possible loss of control of the aircraft.

