AD/CF6/29 Amdt 1 — High Pressure Turbine - Stage 1 Disc Rim Bolt Holes and Stage 2 Disc Rim and Inner Bolt Holes — 13/94

Applicability: General Electric CF6-45/50/-6 series engines.

Requirement: Action in accordance with the technical requirements of the September 1994 corrected issue of FAA AD 94-09-18.

Note: General Electric Service Bulletins 72-1002, 1003, 1057 and 1059 refer.

Compliance: As specified in the FAA AD with a revised effective date of 8 December 1994.

Background: The requirement arises from an uncontained high pressure turbine disc failure which resulted in an aborted take-off. This amendment incorporates changes to two sections but does not constitute a major change in the thrust of the overall AD document.