AD/CF6/30 CF6-80A Stage 1 High Pressure Turbine Disc Inspection 9/94

Applicability: CF6-80A series engines.

Requirement: Action in accordance with the technical requirements of FAA AD 94-07-04.

Note: General Electric Service Bulletin No 72-604 revision 3 refers.

Compliance: As specified in the FAA AD with an effective date of 18 August 1994.

Background: This directive requires an inspection for cracks in the stage 1 high pressure turbine disc rim bolt holes and replacement as necessary with serviceable parts. It originates from an uncontained stage 1 high pressure turbine failure.